

DELTAHAWK

FAA APPROVED
SUPPLEMENTAL AIRPLANE FLIGHT MANUAL
FOR CESSNA MODELS 172N,
SERIAL NUMBERS 17271035 THROUGH 17274009
FOR STC SA-1356GL
&
INSTALLATION INSTRUCTIONS OF
O-320-D2J, D2G, OR D1A 160 HP ENGINE
IN 1979 THROUGH 1980 CESSNA 172N

The Power of Experience



PENN YAN AERO
2499 Bath Road, Penn Yan, New York 14527
U. S. A.

Phone: 315-536-2333, Fax: 315-536-2335
www.PennYanAero.com

Warranty and Disclaimer

Information in this manual is subject to change without notice. No part of it may be reproduced or transmitted in any form or by any means, including electronic or mechanical, for any purpose, without the express written permission of Penn Yan Aero.

Copyright © 2005 Penn Yan Aero
All right reserved.
Produced in the United States of America

Warranty

Penn Yan Aero warrants all equipment manufactured by it and bearing its nameplate and all repairs made by Penn Yan Aero to be free from defects in material or workmanship under normal use and service for a period of one year from the date of delivery to the Purchaser. Penn Yan Aero will repair or replace, at its option, without charge, F.O.B. Penn Yan Aero any equipment or part thereof found to be defective in material or workmanship if such item is returned to Penn Yan Aero transportation prepaid.

Limitation of Warranty and Liability

The liability of Penn Yan Aero (except as to title) shall in no case exceed the cost of correcting defects in products and shall not extend beyond one year from date of delivery. All claims for defective products or parts under this warranty must be made in writing immediately upon discovery. Penn Yan Aero assumes no liability for the consequences of misuse of products by the Purchaser.

The foregoing is expressly in lieu of all other warranties whatsoever, express, implied, or statutory, including without limitation the implied warranties of merchantability and fitness for purpose. Any references by Penn Yan Aero to Purchaser's specifications and similar requirements is only to describe products, and no representation or other terms therein shall have any force or effect. Catalogs, circulars, and similar literature of Penn Yan Aero are issued for general information purposes only and shall not be deemed to modify the provision hereof.

The foregoing is Penn Yan Aero's only obligation and the Purchaser's only remedy for breach of warranty; and except for gross negligence, willful misconduct, and remedies permitted under any other clause of these Terms and Conditions, the foregoing is the Purchaser's only remedy hereunder by way of breach of contract, negligence or otherwise. In no event shall the Purchaser be entitled to incidental, special, or consequential damages. Penn Yan Aero does not authorize any agent or representative to warrant product fitness for any particular use or to make any other warranty, express or implied, or to assume any other liability, except as set forth herein.

FAA APPROVED SUPPLEMENT
TO THE
PILOT'S OPERATING HANDBOOK AND
FAA APPROVED AIRPLANE FLIGHT MANUAL

FOR

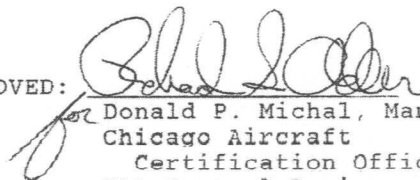
CESSNA MODELS 172N, S/N 17271035 THRU 17274009

REG. NO. N5928E

SER. NO. 17271964

This Supplement must be attached to the Pilot's Operating Handbook (POH) and FAA Approved Airplane Flight Manual when the aircraft is modified by the installation of an O-320-D series engine and the gross weight is increased to 2400 lbs in accordance with STC # SA1356GL. The information contained herein supplements or supersedes the basic manual only in those areas listed. For limitations, procedures and performance information not contained in this supplement, consult the basic POH and FAA Approved Airplane Flight Manual.

FAA APPROVED:



for Donald P. Michal, Manager
Chicago Aircraft
Certification Office
FAA Central Region

DATE: MAR 01 1989

Page 1 of 11

Penn Yan Aero Service, Inc.
2499 Bath Road, Airport
Penn Yan, NY 14527-9599

POH and AFM Supplement
for Cessna 172N

SECTION I - General

DESCRIPTIVE DATA

A. Engine

Number of engines: 1
Engine Manufacturer: Textron Lycoming
Engine Model: O-320-D2J, -D2G, -D1A
Horsepower Rating and Speed: 160 rated BHP at 2700 RPM

SECTION II - Limitations

- A. The following placard must be displayed adjacent to the flap position selector switch:

MAXIMUM FLAP TRAVEL IS 30°

B. C.G. Range

Landplane:

Normal category	(+39.5) to (+47.3) at 2400 lb.
	(+35.0) to (+47.3) at 1950 lb. or less
Utility category	(+36.5) to (+40.5) at 2100 lb.
	(+35.0) to (+40.5) at 1950 lb. or less

Floatplane: (Edo 89-2000 or 89A2000 floats)

Normal category	(+39.8) to (+45.5) at 2220 lb.
	(+36.4) to (+45.5) at 1825 lb. or less

Straight line variation between points given.

SECTION III - Emergency Procedures - No Change.

SECTION IV - Normal Procedures - No Change.

SECTION V - Performance - See pages 3 thru 10.

SECTION VI - Weight and Balance - See Page 11.

FAA APPROVED

DATE: MAR 01 1989

MAR 01 1989

CESSNA
MODEL 172NAircraft Modified
Per Penn Yan STC
2400 lb. gross wt.SECTION 5
PERFORMANCE

required to complete the trip with ample reserve.

LANDING

A procedure similar to takeoff should be used for estimating the landing distance at the destination airport. Figure 5-11 presents landing distance information for the short field technique. The distances corresponding to 2000 feet and 30°C are as follows:

Ground roll	610 Feet
Total distance to clear a 50-foot obstacle	1390 Feet

A correction for the effect of wind may be made based on Note 2 of the landing chart using the same procedure as outlined for takeoff.

DEMONSTRATED OPERATING TEMPERATURE

Satisfactory engine cooling has been demonstrated for this airplane with an outside air temperature 23°C above standard. This is not to be considered as an operating limitation. Reference should be made to Section 2 for engine operating limitations.

AIRSPEED CALIBRATION**NORMAL STATIC SOURCE**

CONDITION:
Power required for level flight or maximum rated RPM dive.

FLAPS UP	50	60	70	80	90	100	110	120	130	140	150	160
KIAS	56	62	70	79	89	98	107	117	126	135	145	154
KCAS												
FLAPS 10°	40	50	60	70	80	90	100	110	120	130	140	150
KIAS	49	55	62	70	79	89	98	108	117	126	135	145
KCAS												
FLAPS 30°	40	50	60	70	80	85	85	85	85	85	85	85
KIAS	47	53	61	70	80	84	84	84	84	84	84	84
KCAS												

Figure 5-1. Airspeed Calibration (Sheet 1 of 2)

SECTION 5
PERFORMANCE

Aircraft Modified
Per Penn Yan STC
2400 lb. gross wt.

CESSNA
MODEL 172N**AIRSPEED CALIBRATION**
ALTERNATE STATIC SOURCE**HEATER/VENTS AND WINDOWS CLOSED**

FLAPS UP	50	60	70	80	90	100	110	120	130	140	141	141
NORMAL KIAS	51	61	71	82	91	101	111	121	131	141	141	141
ALTERNATE KIAS												
FLAPS 10°	40	50	60	70	80	90	100	110	120	130	140	141
NORMAL KIAS	40	51	61	71	81	90	99	108	118	128	138	140
ALTERNATE KIAS												
FLAPS 30°	40	50	60	70	80	85	85	85	85	85	85	85
NORMAL KIAS	38	50	60	70	79	83	83	83	83	83	83	83
ALTERNATE KIAS												

HEATER/VENTS OPEN AND WINDOWS CLOSED

FLAPS UP	40	50	60	70	80	90	100	110	120	130	140	140
NORMAL KIAS	36	48	59	70	80	89	99	108	118	128	138	139
ALTERNATE KIAS												
FLAPS 10°	40	50	60	70	80	90	100	110	120	130	140	140
NORMAL KIAS	38	49	59	69	79	88	97	106	116	126	136	139
ALTERNATE KIAS												
FLAPS 30°	40	50	60	70	80	85	85	85	85	85	85	85
NORMAL KIAS	34	47	57	67	77	81	81	81	81	81	81	81
ALTERNATE KIAS												

WINDOWS OPEN

FLAPS UP	40	50	60	70	80	90	100	110	120	130	140	140
NORMAL KIAS	26	43	57	70	82	93	103	113	123	133	143	143
ALTERNATE KIAS												
FLAPS 10°	40	50	60	70	80	90	100	110	120	130	140	140
NORMAL KIAS	25	43	57	69	80	91	101	111	121	131	141	141
ALTERNATE KIAS												
FLAPS 30°	40	50	60	70	80	85	85	85	85	85	85	85
NORMAL KIAS	25	41	54	67	78	84	84	84	84	84	84	84
ALTERNATE KIAS												

Figure 5-1. Airspeed Calibration (Sheet 2 of 2)

CESSNA Aircraft Modified SECTION 5
MODEL 172N Per Penn Yan STC PERFORMANCE
2400 lb. gross wt.

CRUISE

The cruising altitude should be selected based on a consideration of trip length, winds aloft, and the airplane's performance. A typical cruising altitude and the expected wind enroute have been given for this sample problem. However, the power setting selection for cruise must be determined based on several considerations. These include the cruise performance characteristics presented in figure 5-8, the range profile charts presented in figure 5-9, and the endurance profile charts presented in figure 5-10.

The relationship between power and range is illustrated by the range profile charts. Considerable fuel savings and longer range result when lower power settings are used. For this sample problem, a cruise power of approximately 65% will be used.

The cruise performance chart, figure 5-8, is entered at 8000 feet altitude and 20°C above standard temperature. These values most nearly correspond to the planned altitude and expected temperature conditions. The engine speed chosen is 2500 RPM, which results in the following:

Power	66%
True airspeed	112 Knots
Cruise fuel flow	7.4 GPH

The power computer may be used to determine power and fuel consumption more accurately during the flight.

FUEL REQUIRED

The total fuel requirement for the flight may be estimated using the performance information in figures 5-7 and 5-8. For this sample problem, figure 5-7 shows that a climb from 2000 feet to 6000 feet requires 1.5 gallons of fuel. The corresponding distance during the climb is 10 nautical miles. These values are for a standard temperature and are sufficiently accurate for most flight planning purposes. However, a further correction for the effect of temperature may be made as noted on the climb chart. The approximate effect of a non-standard temperature is to increase the time, fuel, and distance by 10% for each 10°C above standard temperature, due to the lower rate of climb. In this case, assuming a temperature 18°C above standard, the correction would be:

$$\frac{18^{\circ}\text{C} - 10^{\circ}\text{C}}{10^{\circ}\text{C}} = 10\% = 16\% \text{ Increase}$$

SECTION 5 Aircraft Modified CESSNA
PERFORMANCE Per Penn Yan STC MODEL 172N
2400 lb. gross wt.

With this factor included, the fuel estimate would be calculated as follows:

Fuel to climb, standard temperature	1.6
Increase due to non-standard temperature (1.6 × 16%)	0.3
Corrected fuel to climb	1.9 Gallons

Using a similar procedure for the distance to climb results in 12 nautical miles.

The resultant cruise distance is:

Total distance	320
Climb distance	12
Cruise distance	308 Nautical Miles

With an expected 10 knot headwind, the ground speed for cruise is predicted to be:

112
-10
102 Knots

Therefore, the time required for the cruise portion of the trip is:

308 Nautical Miles ÷ 3.0 Hours
102 Knots

The fuel required for cruise is:

3.0 hours × 7.4 gallons/hour = 22.2 Gallons

A 45-minute reserve requires:

45 × 7.4 gallons/hour = 5.6 Gallons
50

The total estimated fuel required is as follows:

Engine start, taxi, and takeoff	1.1
Climb	1.9
Cruise	22.2
Reserve	5.6
Total fuel required	30.8 Gallons

Once the flight is underway, ground speed checks will provide a more accurate basis for estimating the time enroute and the corresponding fuel

SECTION 5
PERFORMANCE

Aircraft Modified
Per Penn Yan STC
2400 lb. gross wt.

CESSNA
MODEL 172N

MAXIMUM RATE OF CLIMB

CONDITIONS:
Flaps Up
Full Throttle

NOTE:
Mixture leaned above 3000 feet for maximum RPM.

WEIGHT LBS	PRESS ALT FT	CLIMB SPEED KIAS	RATE OF CLIMB - FPM			
			-20°C	0°C	20°C	40°C
2400	S.L.	76	805	745	685	625
	2000	75	695	640	590	525
	4000	74	590	535	480	420
	6000	73	485	430	375	320
	8000	72	380	330	275	220
	10,000	71	275	225	175	---
	12,000	70	175	125	---	---

Figure 5-6. Maximum Rate of Climb

SECTION 5
PERFORMANCE

Aircraft Modified
Per Penn Yan STC
2400 lb. gross wt.

CESSNA
MODEL 172N

TAKEOFF DISTANCE 2200 LBS AND 2000 LBS

SHORT FIELD

REFER TO SHEET 1 FOR APPROPRIATE CONDITIONS AND NOTES.

WEIGHT LBS	TAKEOFF SPEED KIAS		PRESS ALT FT	0°C		10°C		20°C		30°C		40°C	
	LIFT OFF	AT 50 FT		GRND ROLL FT	TOTAL FT TO CLEAR 50 FT OBS	GRND ROLL FT	TOTAL FT TO CLEAR 50 FT OBS	GRND ROLL FT	TOTAL FT TO CLEAR 50 FT OBS	GRND ROLL FT	TOTAL FT TO CLEAR 50 FT OBS	GRND ROLL FT	TOTAL FT TO CLEAR 50 FT OBS
2200	49	54	S.L.	650	1195	700	1280	750	1375	805	1470	865	1575
			1000	710	1310	765	1405	825	1510	885	1615	950	1735
			2000	780	1440	840	1545	905	1660	975	1785	1045	1915
			3000	855	1585	925	1705	995	1835	1070	1975	1150	2130
			4000	945	1750	1020	1890	1100	2040	1180	2200	1270	2375
			5000	1040	1945	1125	2105	1210	2275	1305	2465	1405	2665
			6000	1150	2170	1240	2355	1340	2555	1445	2775	1555	3020
			7000	1270	2440	1375	2655	1485	2890	1605	3165	1730	3450
			8000	1410	2760	1525	3015	1650	3305	1785	3630	1925	4005
2000	46	51	S.L.	525	970	565	1035	605	1110	650	1185	695	1265
			1000	570	1060	615	1135	665	1215	710	1295	765	1385
			2000	625	1160	675	1240	725	1330	780	1425	840	1525
			3000	690	1270	740	1365	800	1465	860	1570	920	1695
			4000	765	1400	815	1500	880	1615	945	1735	1015	1865
			5000	830	1545	900	1660	970	1790	1040	1925	1120	2070
			6000	920	1710	990	1845	1070	1990	1150	2145	1235	2315
			7000	1015	1900	1095	2055	1180	2225	1275	2405	1370	2605
			8000	1125	2125	1215	2305	1310	2500	1410	2715	1520	2950

Figure 5-5. Takeoff Distance (Sheet 2 of 2) Page 5 of 11

MAR 01 1989

SECTION 5
PERFORMANCE
Aircraft Modified
Per Penn Yan STC
2400 lb. gross wt.
CESSNA
MODEL 172N

STALL SPEEDS

CONDITIONS:
Power Off

- NOTES:
1. Altitude loss during a stall recovery may be as much as 230 feet.
2. KIAS values are approximate.

MOST REARWARD CENTER OF GRAVITY

WEIGHT LBS	FLAP DEFLECTION	ANGLE OF BANK											
		0°			30°			45°			60°		
		KIAS	KCAS	KIAS	KIAS	KCAS	KIAS	KIAS	KCAS	KIAS	KIAS	KCAS	KCAS
2400	UP	44	51	47	55	52	61	52	61	52	62	72	72
	10°	35	48	38	52	42	57	42	57	49	68	68	68
	30°	33	46	35	49	39	55	39	55	47	65	65	65

MOST FORWARD CENTER OF GRAVITY

WEIGHT LBS	FLAP DEFLECTION	ANGLE OF BANK											
		0°			30°			45°			60°		
		KIAS	KCAS	KIAS	KIAS	KCAS	KIAS	KIAS	KCAS	KIAS	KIAS	KCAS	KCAS
2400	UP	44	52	47	55	52	62	52	62	62	74	74	74
	10°	37	49	40	53	44	58	44	58	52	69	69	69
	30°	33	46	35	49	39	55	39	55	47	65	65	65

Figure 5-3. Stall Speeds

SECTION 5
PERFORMANCE

Aircraft Modified
Per Penn Yan STC
2400 lb. gross wt.
CESSNA
MODEL 172N

TAKEOFF DISTANCE
MAXIMUM WEIGHT 2400 LBS
SHORT FIELD

WEIGHT LBS	TAKEOFF SPEED KIAS	LIFT AT 50 FT	OFF 50 FT	2400	
8000	1755	795	1460	1605	1770
7000	1580	3140	2755	1425	1580
6000	1425	2755	2445	1285	1425
5000	1285	2445	2185	1165	1285
4000	1165	2185	1960	1055	1165
3000	1055	1960	1770	960	1055
2000	875	1605	1580	800	875
1000	795	1460	1390	710	795
1000	1710	1540	1390	1905	1710
1000	1540	1390	1260	1725	1540
1000	1390	1260	1140	1570	1390
1000	1260	1140	1035	1415	1260
1000	1140	1035	940	1260	1140
1000	940	860	860	1115	940
1000	860	795	795	1015	860
1000	795	725	725	925	795
1000	725	660	660	840	725
1000	660	595	595	760	660
1000	595	530	530	685	595
1000	530	465	465	610	530
1000	465	400	400	535	465
1000	400	335	335	460	400
1000	335	270	270	385	335
1000	270	205	205	310	270
1000	205	140	140	235	205
1000	140	75	75	160	140
1000	75	10	10	90	75

- NOTES:
1. Short field technique as specified in Section 4.
2. Prior to takeoff from fields above 3000 feet elevation, the mixture should be leaned to give maximum RPM in a full throttle, static runup.
3. Decrease distances 10% for each 9 knots headwind. For operation with tailwinds up to 10 knots, increase distances by 10% for each 2 knots.
4. For operation on a dry, grass runway, increase distances by 15% of the "ground roll" figure.

CONDITIONS:
Flaps 10°
Full Throttle Prior to Brake Release
Paved, Level, Dry Runway
Zero Wind

Figure 5-5. Takeoff Distance (Sheet 1 of 2)

MAR 01 1989

CESSNA
 MODEL 172N
 Aircraft Modified
 Per Penn Van STC
 2400 lb. gross wt.

TIME, FUEL, AND DISTANCE TO CLIMB

MAXIMUM RATE OF CLIMB

CONDITIONS:

Flaps Up
Full Throttle
Standard Temperature

NOTES:

1. Add 1.1 gallons of fuel for engine start, taxi and takeoff allowance.
2. Mixture leaned above 3000 feet for maximum RPM.
3. Increase time, fuel and distance by 10% for each 10°C above standard temperature.
4. Distances shown are based on zero wind.

CESSNA Aircraft Modified
MODEL 172N Per Penn Yan STC
2400 lb. gross wt.

RANGE PROFILE
45 MINUTES RESERVE
40 GALLONS USABLE FUEL

CONDITIONS:
2400 Pounds
Recommended Lean Mixture for Cruise
Standard Temperature
Zero Wind

NOTE:

This chart allows for the fuel used for engine start, taxi, takeoff and climb, and the distance during climb.

WEIGHT LBS	PRESSURE ALTITUDE FT	TEMP °C	CLIMB SPEED KIAS	RATE OF CLIMB FPM	FROM SEA LEVEL		
					TIME MIN	FUEL USED GALLONS	DISTANCE NM
2400	S.L.	15	76	700	0	0.0	0
	1000	13	76	655	1	0.3	2
	2000	11	75	610	3	0.6	4
	3000	9	75	560	5	1.0	6
	4000	7	74	515	7	1.4	9
	5000	5	74	470	9	1.7	11
	6000	3	73	425	11	2.2	14
	7000	1	72	375	14	2.6	18
	8000	-1	72	330	17	3.1	22
	9000	-3	71	285	20	3.6	26
	10,000	-5	71	240	24	4.2	32
	11,000	-7	70	190	29	4.9	38
	12,000	-9	70	145	35	5.8	47

Figure 5-7. Time, Fuel, and Distance to Climb

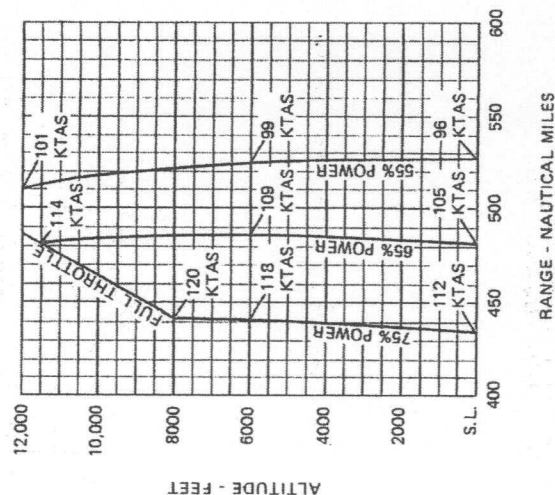


Figure 5-9. Range Profile (Sheet 1 of 3)

MAR 1989

SECTION 5
PERFORMANCE

Aircraft Modified
Per Penn Yan STC
2400 lb. gross wt.

CESSNA
MODEL 172N

CRUISE PERFORMANCE

CONDITIONS:
2400 Pounds
Recommended

Lean Mixture (See Section 4, Cruise)

PRESSURE ALTITUDE FT	RPM	20°C BELOW STANDARD TEMP			STANDARD TEMPERATURE			20°C ABOVE STANDARD TEMP		
		% BHP	KTAS	GPH	% BHP	KTAS	GPH	% BHP	KTAS	GPH
2000	2500	--	--	--	76	114	8.5	72	114	8.1
	2400	65	110	8.1	69	109	7.7	65	108	7.3
	2300	58	104	7.3	62	103	6.9	58	102	6.6
	2200	52	99	6.6	55	97	6.3	53	96	6.1
	2100	52	92	6.0	50	91	5.8	48	89	5.7
4000	2500	--	--	--	76	117	8.5	72	116	8.1
	2400	69	115	8.6	73	114	8.1	69	113	7.7
	2300	62	109	7.8	66	108	7.3	62	107	7.0
	2200	56	104	7.0	60	102	6.6	57	101	6.4
	2100	51	98	6.3	54	96	6.1	51	94	5.9
6000	2500	--	--	--	77	119	8.6	72	118	8.1
	2400	73	114	8.2	69	113	7.8	66	112	7.4
	2300	66	108	7.4	63	107	7.0	60	106	6.7
	2200	60	103	6.7	57	101	6.4	55	99	6.2
	2100	49	96	6.1	52	95	5.9	50	92	5.8
8000	2500	--	--	--	77	121	8.6	73	120	8.1
	2400	77	119	8.7	73	118	8.2	69	117	7.8
	2300	70	113	7.8	65	112	7.4	63	111	7.1
	2200	63	108	7.1	60	106	6.7	58	104	6.5
	2100	57	101	6.4	55	100	6.2	53	97	6.0
10,000	2500	74	118	8.3	70	117	7.8	66	115	7.4
	2400	67	112	7.5	64	111	7.1	61	109	6.8
	2300	61	106	6.8	58	105	6.5	56	102	6.3
	2200	55	100	6.3	53	98	6.0	51	96	5.9
	2100	50	93	5.8	48	91	5.7	47	89	5.6
12,000	2500	67	114	7.5	64	112	7.1	61	111	6.9
	2400	59	111	7.2	61	109	6.8	59	107	6.6
	2300	53	105	6.6	56	103	6.3	54	100	6.1
	2100	48	98	6.1	51	96	5.9	50	94	5.8

Figure 5-8. Cruise Performance

SECTION 5
PERFORMANCE

Aircraft Modified
Per Penn Yan STC
2400 lb. gross wt.

CESSNA
MODEL 172N

RANGE PROFILE
45 MINUTES RESERVE
50 GALLONS USABLE FUEL

CONDITIONS:
2400 Pounds
Recommended Lean Mixture for Cruise
Standard Temperature
Zero Wind

NOTE:
This chart allows for the fuel used for engine start, taxi, takeoff and climb, and the distance during climb.

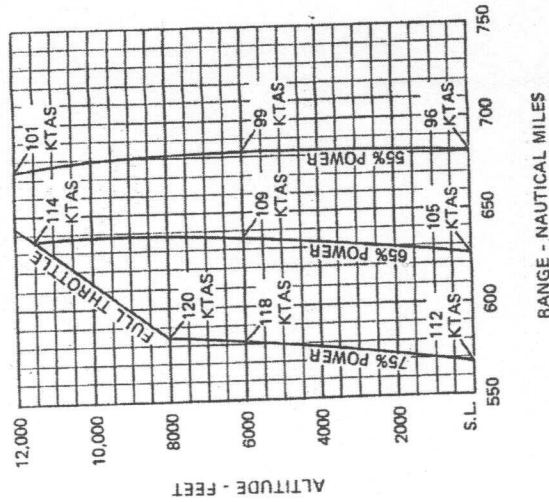


Figure 5-9. Range Profile (Sheet 2 of 3)

MAR 01 1989

SECTION 5
PERFORMANCE

Aircraft Modified
Per Penn Yan STC
2400 lb. gross wt.

CESSNA
MODEL 172N

ENDURANCE PROFILE
45 MINUTES RESERVE
40 GALLONS USABLE FUEL

CONDITIONS:
2400 Pounds
Recommended Lean Mixture for Cruise
Standard Temperature

NOTE:
This chart allows for the fuel used for engine start, taxi, takeoff and climb, and the time during climb.

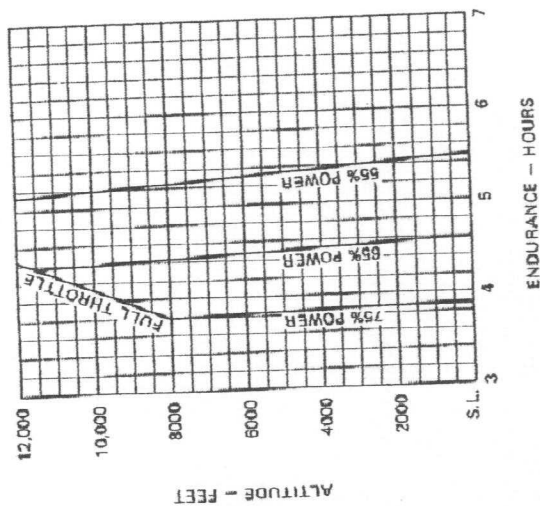


Figure 5-10. Endurance Profile (Sheet 1 of 3)

CESSNA
MODEL 172N

Aircraft Modified
Per Penn Yan STC
2400 lb. gross wt.

SECTION 5
PERFORMANCE

ENDURANCE PROFILE
45 MINUTES RESERVE
50 GALLONS USABLE FUEL

CONDITIONS:
2400 Pounds
Recommended Lean Mixture for Cruise
Standard Temperature

NOTE:
This chart allows for the fuel used for engine start, taxi, takeoff and climb, and the time during climb.

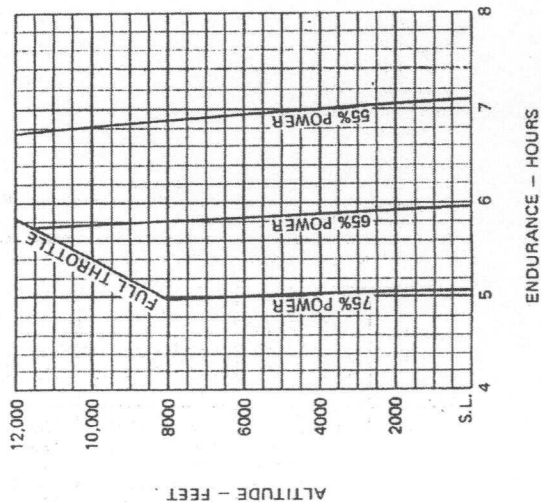


Figure 5-10. Endurance Profile (Sheet 2 of 3)

LANDING DISTANCE

SHORT FIELD

CONDITIONS:
Flaps 30°
Power Off
Maximum Braking
Paved, Level, Dry Runway
Zero Wind

NOTES:

1. Short field technique as specified in Section 4.
2. Decrease distances 10% for each 9 knots headwind. For operation with tailwinds up to 10 knots, increase distances by 10% for each 2 knots.
3. For operation on a dry, grass runway, increase distances by 45% of the "ground roll" figure.
4. If a landing with flaps up is necessary, increase the approach speed by 7 KIAS and allow for 35% longer distances.

WEIGHT LBS	SPEED AT 50 FT KIAS	PRESS ALT FT	0°C		10°C		20°C		30°C		40°C	
			GRND ROLL FT	TOTAL FT TO CLEAR 50 FT OBS	GRND ROLL FT	TOTAL FT TO CLEAR 50 FT OBS	GRND ROLL FT	TOTAL FT TO CLEAR 50 FT OBS	GRND ROLL FT	TOTAL FT TO CLEAR 50 FT OBS	GRND ROLL FT	TOTAL FT TO CLEAR 50 FT OBS
2400	61	S.L.	510	1235	530	1265	550	1295	570	1325	585	1350
		1000	530	1265	550	1295	570	1325	590	1360	610	1390
		2000	550	1295	570	1330	590	1360	610	1390	630	1425
		3000	570	1330	590	1360	615	1395	635	1430	655	1460
		4000	595	1365	615	1400	635	1430	660	1470	680	1500
		5000	615	1400	640	1435	660	1470	685	1510	705	1540
		6000	640	1435	660	1470	685	1510	710	1550	730	1580
		7000	665	1475	690	1515	710	1550	735	1590	760	1630
		8000	690	1515	715	1555	740	1595	765	1635	790	1675

Figure 5-11. Landing Distance

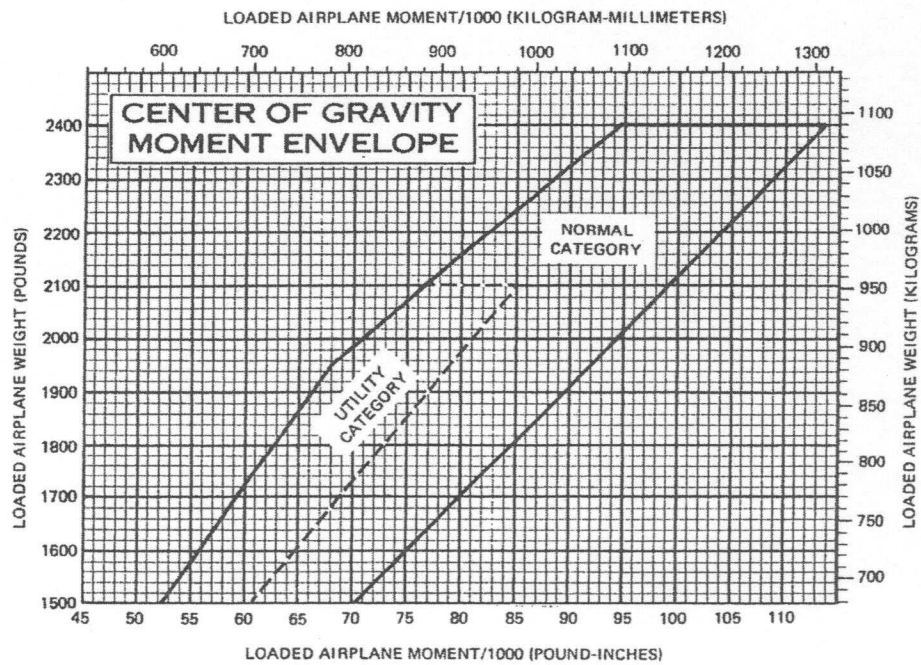


Figure 6-7. Center of Gravity Moment Envelope

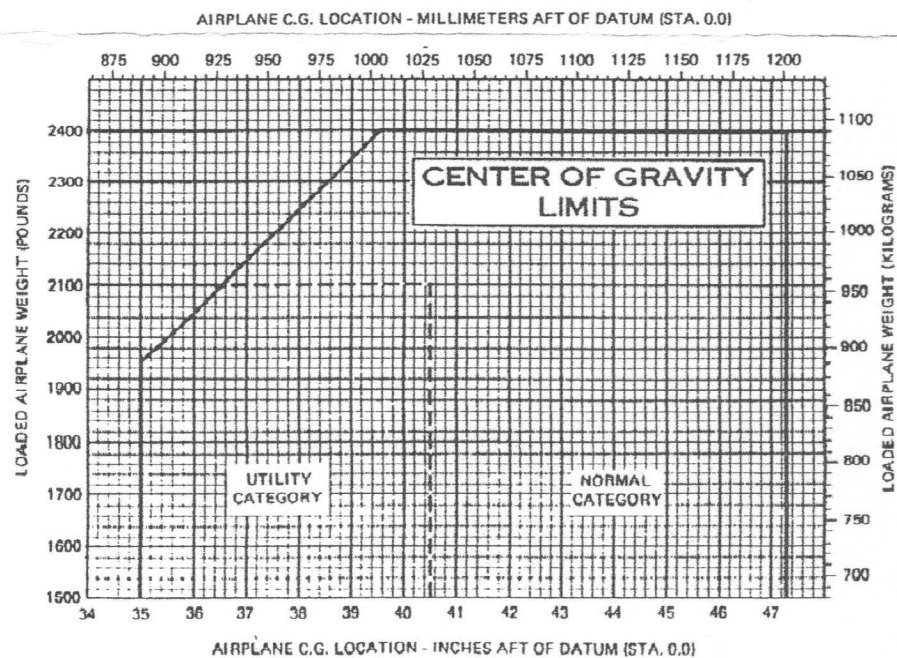


Figure 6-8. Center of Gravity Limits

Mid Island Air Service
101 Hering Drive
Ronkonkoma NY 11779

MODEL: 172N

REGISTRATION: N5928E

[illegible]

* ITEM NUMBERS WHEN LISTED IN THE PERTINENT AIRCRAFT SPECIFICATION MAY BE USED IN LIEU OF "ITEM, MAKE, AND MODEL"

Useful load: 894.6

Prepared By

Date _____

3/9/2018

Mid Island Air Service
101 Hering Drive
Ronkonkoma NY 11779

REGISTRATION: N5928E

[illegible]

* ITEM NUMBERS WHEN LISTED IN THE PERTINENT AIRCRAFT SPECIFICATION MAY BE USED IN LIEU OF "ITEM, MAKE, AND MODEL"

Useful load: 894.6

Prepared By

Date 3/9/2018